

- had an equation of motion for \ddot{r}
- recast in terms of new variable $u = \frac{1}{r}$
- changed time derivatives to φ derivatives
- goal: $u(\varphi) = \frac{1}{r(\varphi)}$ polar eqn of orbit

$$\frac{d^2 u}{d\varphi^2} = -u(\varphi) - \frac{\mu}{l^2 [u(\varphi)]^2} F \quad \text{where } F(u) \text{ is the force law}$$

- clearly if $F \propto u^2 = \frac{1}{r^2}$ the eqn is simple,
form $u'' = -u + C$ choice of origin eliminates C
 \Rightarrow SHM eqn!

$$\text{So let } F = \frac{-\gamma}{r^2} = -\gamma u^2 \Rightarrow \frac{d^2 u}{d\varphi^2} = -u(\varphi) - \underbrace{\frac{\gamma \mu}{l^2}}_{\text{const}}$$

now shift origin: let $W(\varphi) = u(\varphi) - \frac{\gamma \mu}{l^2}$, $w'' = u'' \Rightarrow \underline{w'' = -w}$

$$\Rightarrow W(\varphi) = A \cos(\varphi - \delta) \quad \text{known soln!}$$

$$\Rightarrow u(\varphi) = A \cos(\varphi - \delta) + \frac{\gamma \mu}{l^2} = \frac{\gamma \mu}{l^2} \left(1 + \frac{A l^2}{\gamma \mu} \cos(\varphi - \delta) \right)$$

- define $\epsilon = \frac{A l^2}{\gamma \mu}$, $c = \frac{l^2}{\gamma \mu}$

- choose φ origin such that $\delta = 0$

$$\Rightarrow u(\varphi) = \frac{1}{c} (1 + \epsilon \cos \varphi) \Rightarrow \boxed{r(\varphi) = \frac{c}{1 + \epsilon \cos \varphi}}$$

note $c = r(\varphi = 90^\circ)$

for gravity, $\gamma = G m_1 m_2 = G \mu M$

$$\Rightarrow c = \frac{l^2}{G \mu^2 M} = [\text{distance}] \quad \epsilon = \frac{A l^2}{G \mu^2 M} = [\text{dimensionless}]$$

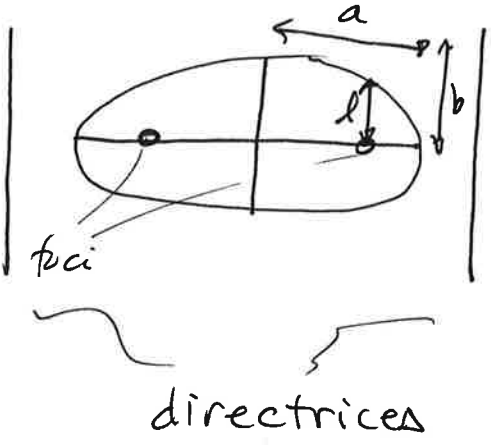
What are these curves?

recall $r = \frac{l}{1 + e \cos \varphi}$ = polar expr for conic sections!

$e = \text{eccentricity} = \begin{cases} 0 & \text{circle} \\ 0 < e < 1 & \text{ellipse} \\ e = 1 & \text{parabola} \\ e > 1 & \text{hyperbola} \end{cases}$

$l = \text{semilatus rectum}$

$\left. \begin{matrix} \text{bound, closed} \\ \text{unbound} \end{matrix} \right\}$



$e < 1$

$a = \text{semimajor axis}$

$b = \text{semiminor axis}$

$e = \sqrt{1 - b^2/a^2}$

$l = \text{semilatus rectum} = \frac{b^2}{a}$

= 1/2 of chord through focus parallel to directrix

= $l^2 / G M^2$

Bounded orbits

forget we identified e and $c = l$
what are the properties of the orbits for $0 \leq e < 1$?

$r = \frac{e}{1 + e \cos \varphi}$

- for $e < 1$, the denominator never vanishes
 - so $r(\varphi)$ is bounded for all φ
 - due to $\cos \varphi$ term, $r(0) = r(2\pi)$
orbits are closed in 1 revolution

- if $e \geq 1$ denominator vanishes for some φ
so $r \rightarrow \infty$ as φ approaches that angle - unbounded!

• $e = 1$ is the boundary btw closed & unclosed orbits

with $e < 1$, $\cos \varphi$ has min/max of ± 1 , so $r(\varphi)$ oscillates between

$r_{\min} = \frac{e}{1 + e}$ $r_{\max} = \frac{e}{1 - e}$

perihelion, $\varphi = 0$ aphelion, $\varphi = \pi$

What is the shape, if we didn't recognize polar form of conic sections?
(long detour)

note $r \cos \phi = x$, and $r(1 + e \cos \phi) = c$

$\Rightarrow r(1 + e \cos \phi) = r(1 + ex) = c$ or $r = c - ex$

$r^2 = x^2 + y^2 = (c - ex)^2 = c^2 - 2cex + e^2x^2$

$\Rightarrow c^2 = (1 - e^2)x^2 + 2cex + y^2$ divide by $1 - e^2$

$\frac{c^2}{1 - e^2} = x^2 + \frac{ce}{1 - e^2}(2x) + \frac{y^2}{1 - e^2}$ let $d = \frac{ce}{1 - e^2}$

$\frac{c^2}{1 - e^2} = x^2 + 2dx + \frac{y^2}{1 - e^2}$

- looks like $x^2 + 2dx + d^2 \dots$
- Complete square by adding d^2 to both sides

$x^2 + 2dx + d^2 + \frac{y^2}{1 - e^2} = \frac{c^2}{1 - e^2} + \frac{c^2 e^2}{1 - e^2}$

$(x + d)^2 + \frac{y^2}{1 - e^2} = \frac{c^2}{1 - e^2} \left(1 + \frac{e^2}{1 - e^2}\right) = \frac{c^2}{1 - e^2} \left(\frac{1 - e^2 + e^2}{1 - e^2}\right) = \frac{c^2}{(1 - e^2)^2}$

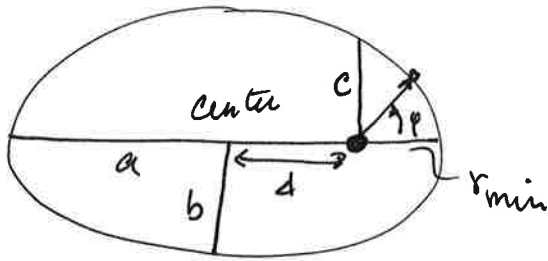
now let $a = \frac{c}{1 - e^2}$ and note $d = ae$

$(x + d)^2 + \frac{y^2}{1 - e^2} = a^2 \Rightarrow \frac{(x + d)^2}{a^2} + \frac{y^2}{a^2(1 - e^2)} = 1$

now let $b = a\sqrt{1 - e^2}$

$\Rightarrow \frac{(x + d)^2}{a^2} + \frac{y^2}{b^2} = 1$

ellipse in cartesian coord!
bound orbits for $e < 1$ are ellipses; $e = 0$ special case is a circle



$$\frac{(x+d)^2}{a^2} + \frac{y^2}{b^2} = 1$$

a = semimajor axis
 b = semiminor axis

orbit is an ellipse with one body at a focus

$$a+d = r_{\max}$$

$$c = \text{Semilatus rectum} = r(\phi=90^\circ) = \frac{l^2}{\mu r}$$

$$a = \frac{c}{1-\epsilon^2} \quad b = \frac{c}{\sqrt{1-\epsilon^2}} \quad d = a\epsilon$$

$$\frac{b}{a} = \sqrt{1-\epsilon^2} \quad \leftarrow \text{one definition of eccentricity}$$

Orbital period

recall $\frac{dA}{dt} = \frac{l}{2\mu} = \text{constant}$

and for an ellipse $A = \pi ab$

period is then $\tau = \frac{A}{dA/dt}$ (like $t = \frac{x}{\dot{x}}$ when $\ddot{x} = 0$)

$$\tau = \frac{\pi ab}{l/2\mu} = \frac{2\pi ab\mu}{l}$$

Square, note $b^2 = a^2(1-\epsilon^2)$

$$\tau^2 = \frac{4\pi^2 a^2 b^2 \mu^2}{l^2} = \frac{4\pi^2 a^4 (1-\epsilon^2) \mu^2}{l^2} = 4\pi^2 \frac{a^3 \mu^2}{l^2} \quad c = a(1-\epsilon^2)$$

$$\text{but } c = \frac{l^2}{\mu r}$$

$$\Rightarrow \tau^2 = 4\pi^2 \frac{a^3 \mu}{\gamma}$$

$$\Rightarrow \tau^2 = 4\pi^2 \frac{a^3}{GM}$$

for gravity, $\gamma = Gm_1 m_2 = G\mu M$

but for a system like the earth-sun,
 $M \approx M_s$ earth mass negligible

$$\Rightarrow \tau^2 = \frac{4\pi^2 a^3}{GM_s}$$

• Same for any body orbiting the sun

• Kepler's 3rd law $\tau^2 \propto a^3$

• known τ, a , indep measurement of G
 allows determination of solar mass!

Same for satellites of earth. In low earth orbit, height above surface $h \ll$ radius of earth R_e

low earth orbit, $a \approx R_e = 6.38 \times 10^6 \text{ m}$

$$\tau = 2\pi \sqrt{\frac{R_e^3}{GM_e}} = 2\pi \sqrt{\frac{R_e}{g}} \approx 85 \text{ min}$$

low earth orbits are all $\sim 1\frac{1}{2}$ hours

Eccentricity & Energy

• at closest approach, r_{\min} , $E = \mathcal{U}(r_{\min})$ and $T = 0$

$$E = \mathcal{U}_{\text{eff}}(r_{\min}) = -\frac{\gamma}{r_{\min}} + \frac{l^2}{2\mu r_{\min}^2} = \frac{1}{2r_{\min}} \left(\frac{l^2}{\mu r_{\min}} - 2\gamma \right)$$

$$\text{note } r_{\min} = \frac{c}{1+E} = \frac{l^2}{\mu\gamma(1+E)}$$

$$\Rightarrow E = \frac{\mu\gamma(1+E)}{2l^2} [\gamma(1+E) - 2\gamma] = \frac{\mu\gamma^2}{2l^2} \underbrace{[(1+E)^2 - 2(1+E)]}_{E^2 - 1}$$

$$\Rightarrow \boxed{E = \frac{\gamma^2 \mu}{2l^2} (E^2 - 1)}$$

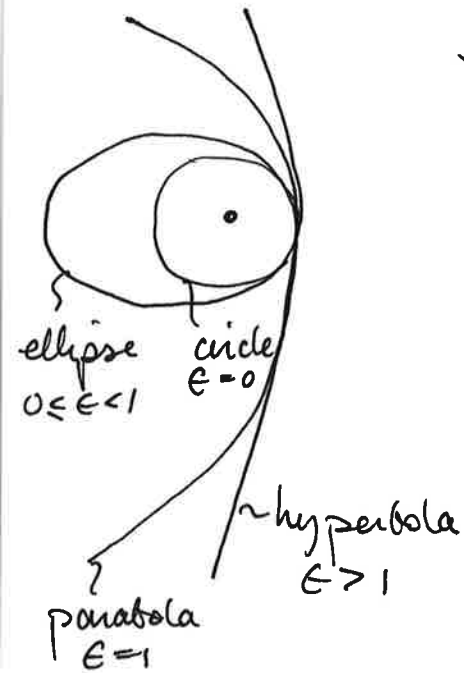
$E < 1$ (bounded; circle or ellipse)

$\Rightarrow E < 0$ also cond. for bound

$E \geq 1$ $E \geq 0$ unbounded

! for a given value of l , lowest energy orbit is circular

Unbounded orbits



$$r = \frac{e}{1 + e \cos \varphi} \quad E \geq 0, e \geq 1$$

• if $e \geq 1$, $r(\varphi) \rightarrow \infty$ for some φ

• at $e = 1$, it is for $\varphi = \pm \pi$

so $r \rightarrow \infty$ as $\varphi \rightarrow \pm \pi$

\Rightarrow unbounded - too high T to bind

• cartesian $y^2 = c^2 - 2cx$ for $e = 1$, parabola

• general $e > 1$ $\frac{(x-s)^2}{\alpha^2} - \frac{y^2}{\beta^2} = 1$ hyperbola

$$\alpha = \frac{e}{e^2 - 1} = -a \quad \beta = \frac{c}{\sqrt{e^2 - 1}} = \frac{b}{i}$$

$$s = e\alpha = -ae = -d$$